

Technical Standard Orders for Aircraft Materials,
Parts, Processes, and Appliances

Part 514 contains minimum performance standards and specifications of materials, parts, processes, and appliances used in aircraft and implements the provisions of sections 3.18, 4a.31, 4b.18, 6.18 and 7.18 of the Civil Air Regulations. The regulation uses the Technical Standard Order system which, in brief, provides for FAA-industry cooperation in the development of performance standards and specifications which are adopted by the Administrator as Technical Standard Orders, and a form of self-regulation by industry in demonstrating compliance with these orders.

Part 514 consists of two subparts. Subpart A contains the general requirements applicable to all Technical Standard Orders. These provisions are summarized below for the convenient reference of the public. Subpart B contains the technical standards and specifications to which a particular product must conform, and each Technical Standard Order is set forth in the appropriate section of Subpart B. The subject Technical Standard Order is printed below. ANY TECHNICAL STANDARD ORDER MAY BE OBTAINED BY SENDING A REQUEST TO FAA, WASHINGTON 25, D. C.

SUBPART A--GENERAL

This subpart provides, in part, that a manufacturer of an aircraft material, part, process, or appliance for which standards are established in Subpart B, prior to its distribution for use on a civil aircraft of the United States, shall furnish a written statement of conformance certifying that the material, part, process, or appliance meets the applicable performance standards established in this part. The statement of conformance must be signed by a person duly authorized by the manufacturer, and furnished to the Chief, Engineering and Manufacturing Division, Bureau of Flight Standards, Federal Aviation Agency, Washington 25, D. C.

Subpart A also requires appropriate marking of materials, parts, processes, and appliances as follows:

- (a) Name and address of the manufacturer responsible for compliance,
- (b) Equipment name, or type or model designation,
- (c) Weight to the nearest pound and fraction thereof,
- (d) Serial number and/or date of manufacture, and
- (e) Applicable Technical Standard Order (TSO) number.

In addition, Subpart A provides that no deviation will be granted from the performance standards established in Subpart B, and that the Administrator may take appropriate action in the event of noncompliance with Part 514.

for installation on civil aircraft on or after April 1, 1959, shall meet the standards set forth in SAE Aeronautical Standard AS-397A, "Direction Instrument, Non-Magnetic, Stabilized Type (Directional Gyro)," dated July 15, 1958,^{1/} with the exceptions listed in subparagraph (2) of this paragraph. Direction instruments, non-magnetic gyro-stabilized type (directional gyro) approved by the Administrator prior to April 1, 1959, may continue to be manufactured under the provisions of their original approval.

(2) Exceptions. (i) Conformance with the following sections is not required: 3.1; 3.1.1; 3.1.2; 3.2; 4.3.3.

(ii) Substitute the following for section 7.: "Performance tests: The following tests, in addition to any others deemed necessary by the manufacturer, shall be the basis for determining compliance with the performance requirements of this standard."

(b) Marking. In lieu of the weight specified in paragraph (c) of §514.3, the following shall be shown:

(1) Instrument type (I or II)

(2) Rating if applicable, i.e., electrical, vacuum, etc.

(c) Data requirements. One copy each of the manufacturer's operating instructions, schematic diagrams, and installation procedures shall be furnished the Chief, Engineering and Manufacturing Division, Federal Aviation Agency, Washington 25, D. C., with the statement of conformance.

(d) Effective date. April 1, 1959.

^{1/}Copies may be obtained from the Society of Automotive Engineers, Inc., 485 Lexington Ave., New York 17, New York.

<u>in Airframe</u>	<u>Per Sec.</u>	<u>Amplitude (Inches)</u>	<u>Acceleration</u>
Power Plant Mounted	5 - 150	0.100	20g
Wings and Empennage	5 - 500	0.036	10g
Fuselage	5 - 500	0.036	5g
Panel or Rack (with Shockmounts)	5 - 50	0.020	1.5g

3.3.4 Altitude: The instrument shall function and shall not be adversely affected when subjected to a pressure and temperature range equivalent from -1000 feet to 40,000 feet standard altitude per NACA Report 1235 except as limited by application of Paragraph 3.3.1. The instrument shall not be adversely affected when subjected to a pressure of 50 in. Hg. absolute.

3.4 Radio Interference: The instrument shall not be the source of objectionable interference, under operating conditions at any frequencies used on aircraft, either by radiation or feedback, in radio equipment installed in the same aircraft as the instrument.

3.5 Magnetic Effect: The magnetic effect of the instrument shall not adversely affect the operation of other instruments installed in the same aircraft.

4. DETAIL REQUIREMENTS:

4.1 Indicating Method: One of the following methods of indication shall be employed:

Method I - Horizontal drum dial with fixed lubber's line.
Graduations move to the right for right turns.

Method II - Rotating vertical dial with fixed lubber's line at the top. Dial rotates counterclockwise for right turns.

Method III - Rotating pointer with fixed graduated dial. Pointer rotates clockwise for right turns.

4.2 Operating Limits: The instrument shall indicate heading throughout the 360 degree scale range. During dives, climbs or banks up to and at least 55 degrees displacement from level flight the instrument shall remain functional; however, the heading error involved through the gimbal system need not be corrected.

4.3 Dial Markings:

- 4.3.2 Visibility: Index and dial markings shall be visible from any point within the frustum of a cone the side of which makes an angle of 30 degrees with the perpendicular to the dial and the small diameter of which is the aperture of the instrument case. At least two numerals shall be simultaneously visible.
- 4.3.3 Finish: Unless otherwise specified by the user, matte white material shall be applied to major graduations, numerals and pointers. Non-functional surfaces shall be a durable dull black.
- 4.4 Course Setting Provisions: A means shall be provided for manually setting the directional indicator dial (or pointer) indication to any heading desired.
- 4.5 Gyro Caging Provisions: Unless the gyro assembly has unrestricted freedom of operation in the pitch and roll axes, means shall be provided for caging and/or releveing the gyro. Means shall be provided to indicate when the gyro is caged, except when it is not possible to leave the gyro in caged condition.
- 4.6 Power Malfunction Indication: Means shall be incorporated in the instrument to indicate when adequate power (voltage and/or current) is not being made available to all the phases required for the proper operation of the instrument. The indicating means shall indicate a failure or a malfunction in a positive manner.

5. TEST CONDITIONS:

- 5.1 Atmospheric Conditions: Unless otherwise specified, all tests required by this Aeronautical Standard shall be conducted at an atmospheric pressure of approximately 29.92 inches of mercury and at an ambient temperature of approximately 25C and at a relative humidity of not greater than 85 percent. When tests are made with the atmospheric pressure or the temperature substantially different from these values, allowance shall be made for the variation from the specified conditions.
- 5.2 Vibration (to minimize friction): Unless otherwise specified, all tests for performance may be conducted with the instrument subjected to a vibration of 0.002 to 0.005 inch double amplitude at a frequency of 1500 to 2000 cycles per minute. The term double amplitude as used herein indicates the total displacement from positive maximum to negative maximum.
- 5.3 Vibration Equipment: Vibration equipment shall be used which will provide frequencies and amplitudes consistent with the requirements of Paragraph 3.3.3 with the following characteristics:

on the instrument case will describe a circle in a plane inclined 45 degrees to the horizontal plane, the diameter of which is equal to the double amplitude specified.

5.4 Power Conditions: Unless otherwise specified all tests shall be conducted at the power rating recommended by the manufacturer.

5.5 Position: Unless otherwise specified, all tests shall be made with the instrument (indicators, amplifiers, transmitters, etc.) mounted in their normal operating position.

6. INDIVIDUAL PERFORMANCE REQUIREMENTS: All instruments shall be subjected to whatever tests the manufacturer deems necessary to demonstrate specific compliance with this Aeronautical Standard including the following requirements where applicable.

6.1 Type I Requirements:

6.1.1 Starting: The gyro rotor shall start to rotate and continue to run on a suction not to exceed 50 percent of rated value. Rated instrument performance speed shall be reached within two minutes after normal rated suction is applied.

6.1.2 Roll, Pitch and Yaw: The instrument shall be mounted on a test platform which is adjusted to oscillate in roll, pitch and yaw, with a total amplitude of 3 degrees about each axis, at a frequency of 5 to 7 oscillations per minute. With the platform level, and the gyro operating at equilibrium speed and uncaged, the dial (or pointer) reading shall be noted. The platform shall then be started in its roll, pitch and yaw movement. At the end of a ten minute period the oscillation shall be stopped, the platform realigned to its starting position, and the instrument dial (or pointer) reading noted. The amount of drift of the dial (or pointer) in either direction during the ten minute test period shall not exceed 4 degrees.

6.1.3 Heading Stability: The instrument shall be mounted on a turn table, tilted 54 (+1) degrees from the vertical and the reading noted. The turn table shall be rotated one complete revolution about its vertical axis at 360 (+30) degrees per minute and the drift of the dial (or pointer) shall not exceed two degrees. The test shall be repeated rotating the turn table in the opposite direction.

6.2 Type II Requirements:

voltage.

6.2.2 Roll, Pitch and Yaw: Same as for Type I.

6.2.3 Heading Stability: Same as for Type I.

6.3 Dielectric: Ungrounded instruments or grounded instruments prior to connection of internal ground wire, shall be tested by either the method of inspection of Paragraph 6.3.1 or 6.3.2.

6.3.1 Insulation Resistance: The insulation resistance measured at 500 volts D. C. (200 volts for hermetically sealed, inert gas filled instruments) between all electrical circuits connected together and the metallic case shall not be less than 20 megohms.

6.3.2 Dielectric Strength: The insulation shall withstand without evidence of damage the application of a sinusoidal voltage at a commercial frequency between all electrical circuits connected together and the metallic case, for a period of five seconds. The RMS value of the sinusoidal voltage applied shall be either five (5) times the maximum instrument operating voltage, or 500 volts, whichever is lower, except that on hermetically sealed (inert gas filled instruments) the test voltage shall be 200 volts RMS.

6.3.2.1 Instruments having a permanent internal ground connection shall be tested as follows:

The insulation shall withstand without evidence of damage the application of a sinusoidal voltage at a commercial frequency between each electric circuit and the metallic case, for a period of five (5) seconds. The RMS value of the sinusoidal voltage applied shall be 1.25 times the maximum circuit operating voltage obtainable between the test points.

7. QUALIFICATION TESTS: As many instruments as deemed necessary to demonstrate that all instruments will comply with the requirements of this section shall be tested in accordance with the manufacturer's recommendations.

7.1 Low Temperature: The instruments, or components shall be subjected to the temperature indicated in the following table in accordance with their location in the aircraft. After exposure to this temperature for five hours, the instrument shall start upon application of rated power. The amount of drift of the dial (or pointer) in either direction during a ten minute period shall not exceed five degrees.

<u>Instrument Location</u>	<u>Temperature</u>
Heated Area (Temperature Controlled)	- 30 C
Unheated Area (Temperature Uncontrolled)	- 55 C

- 7.3 Extreme Temperature Exposure: The instrument, or components, shall, after alternate exposures to ambient temperature of -65°C and 70°C for periods of 24 hours each and a delay of 3 hours at room temperature following completion of the exposure, meet the requirements of Section 6 at room temperature. There shall be no evidence of damage as a result of exposure to the extreme temperatures specified herein.
- 7.4 Magnetic Effect: The magnetic effect of the Indicator shall be determined in terms of the deflection of a free magnet, approximately $1\frac{1}{2}$ inches long, in a magnetic field with a horizontal intensity of 0.18 (± 0.01) gauss when the indicator is held in various positions on an east-west line with its nearest part 12 inches from the center of the magnet. This test shall first be made with the indicator not operating and then shall be repeated with the indicator in normal operation. The maximum deflection of the free magnet shall not exceed 5 degrees for any pointer or dial position.
- 7.5 Humidity: The instrument (indicator, remote gyro, amplifier, etc.) shall be mounted in a chamber maintained at a temperature of $70 \pm 2^{\circ}\text{C}$ and a relative humidity of $95 \pm 5\%$ for a period of six hours. After this period the heat shall be shut off and the instrument shall be allowed to cool for a period of 18 hours in this atmosphere in which the humidity rises to 100% as the temperature decreases to not more than 38°C . This complete cycle shall be conducted:
- a. Five times for components located in uncontrolled temperature areas.
 - b. Once for components located in controlled temperature areas.

Immediately after recycling; there shall be no evidence of damage or corrosion, which affects performance, following this test, and the instrument shall meet the requirements of Section 6 except for Paragraph 6.3.

the parts. The amplitude used may be any convenient value that does not exceed the maximum double amplitude and the maximum acceleration specified in Paragraph 3.3.3.

The instrument shall then be subjected to a vibration at the appropriate maximum double amplitude or maximum acceleration specified in Paragraph 3.3.3 at the resonant frequency for a period of one hour in each axis or with circular motion vibration, whichever is applicable. When more than one resonant frequency is encountered with vibration applied along any one axis, a test period may be accomplished at the most severe resonance, or the period may be divided among the resonant frequencies, whichever shall be considered most likely to produce failure. The test period shall not be less than one-half hour at any resonant mode. When resonant frequencies are not apparent within the specified frequency range, the instrument shall be vibrated for two hours in accordance with the vibration requirements schedule (Paragraph 3.3.3) at the maximum double amplitude and the frequency to provide the maximum acceleration.

Cycling - The instrument, while operating, shall be tested with the frequency cycles between limits specified in Paragraph 3.3.3 in 15 minute cycles for a period of one hour in each axis at an applied double amplitude specified in Paragraph 3.3.3 or an acceleration specified in Paragraph 3.3.3 whichever is the limiting value or a total of three hours for circular motion vibration, whichever is applicable.

The instrument shall be tested during vibration to determine that it is functioning properly. After the completion of this vibration test, no damage shall be evident and the instrument shall meet the requirements of Section 6 (other than Paragraph 6.3).

